

UNCLASSIFIED

AD NUMBER

**AD820805**

NEW LIMITATION CHANGE

TO

**Approved for public release, distribution  
unlimited**

FROM

**Distribution authorized to U.S. Gov't.  
agencies and their contractors;  
Administrative/Operational Use; SEP 1967.  
Other requests shall be referred to Air  
Force Weapons Lab., Kirtland AFB, NM  
87117.**

AUTHORITY

**AFWL ltr, 30 Nov 1971**

THIS PAGE IS UNCLASSIFIED

**UNCLASSIFIED**

NOTICE: When government or other drawings, specifications or other data are used for any purpose other than in connection with a definitely related government procurement operation, the U. S. Government thereby incurs no responsibility, nor any obligation whatsoever; and the fact that the Government may have formulated, furnished, or in any way supplied the said drawing, specifications, or other data is not to be regarded by implication or otherwise as in any manner licensing the holder or any other person or corporation, or conveying any rights or permission to manufacture, use or sell any patented invention that may in any way be related thereto.

AD 820805

**DEFENSE DOCUMENTATION CENTER**

FOR

**SCIENTIFIC AND TECHNICAL INFORMATION**

CAMERON STATION ALEXANDRIA VIRGINIA



**UNCLASSIFIED**

Reproduced by  
**NATIONAL TECHNICAL**  
**INFORMATION SERVICE**  
Springfield, Va 22151

This document has been approved  
for public release and sale.  
Reproduction for distribution

54

UNCLASSIFIED

DDC REPORT BIBLIOGRAPHY SEARCH CONTROL NO. /FAYE2

AD-820 805 20/11  
STANFORD RESEARCH INST MENLO PARK CALIF POULTER LABS

EQUATION OF STATE OF SOLIDS. II. ALUMINUM AND  
TEFLON. (U)

DESCRIPTIVE NOTE: TECHNICAL REPT. 6 MAY 66-6 MAY 67,  
SEP 67 52P ANDERSON, G. D. ;FAHRENBRUCH,  
A. L. ;  
CONTRACT: AF 29(601)-7214  
PROJ: AF-5710, SRI-FGU-6047  
MONITOR: AFWL TR-67-43

UNCLASSIFIED REPORT  
DISTRIBUTION: NO FOREIGN WITHOUT APPROVAL OF AIR  
FORCE WEAPONS LAB., ATTN: WLRP, KIRLAND  
AFB, N. MEX 87117.

DESCRIPTORS: (\*EQUATIONS OF STATE, \*ALUMINUM),  
(\*HALOCARBON PLASTICS, EQUATIONS OF STATE),  
SHOCK WAVES, PRESSURE, VOLUME, STRESSES,  
EXPLOSION EFFECTS, POWDER METALS, PROJECTILES,  
VELOCITY, HELIUM, CODING, COMPUTERS, SOLIDS,  
TARGETS (U)

IDENTIFIERS: ALUMINUM ALLOY 2024, GAS GUNS,  
HUGONIOTS, TEFLON (U)

THE PRESSURE-VOLUME-ENERGY (P-V-E) EQUATION  
OF STATE OF ALUMINUM AND TEFLON HAS BEEN  
INVESTIGATED. THE P-V-E EQUATION OF STATE OF  
A MATERIAL IS NEEDED TO SOLVE NONREACTIVE FLOW  
PROBLEMS USING COMPUTER CODES SUCH AS PUFF.  
EXPLOSIVELY INDUCED SHOCK WAVES HAVE BEEN USED TO  
GENERATE HUGONIOT CURVES FOR SOLID AND POROUS  
SPECIMENS OF ALUMINUM AND TEFLON. FOR ALUMINUM  
IT IS FOUND THAT THE FOLLOWING P-V-E EQUATION  
OF STATE CAN REPRODUCE THE EXPERIMENTAL DATA QUITE  
NICELY:  $P = BE + G(V)$  B IS A CONSTANT AND  
G IS AN ARBITRARY FUNCTION OF V. THIS  
CORRESPONDS TO  $\Gamma = \text{CONSTANT}$ , WHERE  $\Gamma$  IS  
GRUNEISEN'S RATIO. BECAUSE OF THE SPREAD OF THE  
TEFLON DATA IN THE P-V PLANE, IT WAS NOT  
POSSIBLE TO OBTAIN A P-V-E FIT. HOWEVER, THE  
DATA IMPLY THAT GRUNEISEN'S RATIO VARIES FROM ABOUT  
0.7 TO 1.4 AT HIGH PRESSURES. GAS GUN EXPERIMENTS  
USING MANGANIN WIRE PRESSURE RECORDINGS WERE  
PERFORMED. SOUND SPEEDS DEDUCED FROM THESE RECORDS  
IMPLY THAT TEFLON BECOMES ANOMALOUSLY STIFF UNDER  
SHOCK. (AUTHOR) (U)

UNCLASSIFIED

/FAYE2

AFWL-TR-67-43

(20)

EQUATION OF STATE OF SOLIDS  
II  
ALUMINUM AND TEFILON

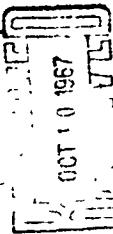
G. D. Anderson  
A. L. Fahrenbruch

Stanford Research Institute  
Poulier Laboratories  
Menlo Park, California 94025  
Contract AF 29(601)-7214

TECHNICAL REPORT NO AFWL-TR-67-43

September 1967

AIR FORCE WEAPONS LABORATORY  
Research and Technology Division  
Air Force Systems Command  
Kirtland Air Force Base  
New Mexico



OCT 10 1967

UTL

AD820805 FILE COPY

11/17  
AFWL-TR-67-43

AFWL-TR  
67-43

EQUATION OF STATE OF SOLIDS

II

ALUMINUM AND TEFILON

G. D. Anderson

A. L. Fahrenbruch

Stanford Research Institute  
Poulier Laboratories  
Menlo Park, California 94025  
Contract AF 29(601)-7214

TECHNICAL REPORT NO AFWL-TR-67-43

This document is subject to special  
export control and may be  
transmitted to foreign governments  
or foreign nationals only  
with prior approval of the  
U.S. Air Force, Kirtland Air  
Force Base, New Mexico, 87117.  
Distribution is limited  
to those persons whose  
knowledge is necessary to  
the performance of their  
duties.

11

## FOREWORD

This report was prepared by the Poulter Laboratories, Stanford Research Institute, Menlo Park, California, under Contract AF 29(601)-7214. The research was performed under Program Element 616.46.01D, Project 510, Subtask 15.018, and was funded by the Defense Atomic Support Agency (DASA).

Inclusive dates of research were 6 May 1966 to 6 May 1967. The report was submitted 11 July 1967 to the Air Force Weapons Laboratory Project Officer, Captain Raymond J. Lawrence, Jr. (WLR).

The project supervisor was Dr. G. D. Anderson and the project leader was Mr. A. L. Fahrenbruch. The authors wish to thank Drs. D. G. Dorian and M. Cooper-thainite for their contributions through discussions on equations of state, and Miss B. Y. Loo for shot analysis and computations.

Contractor's report number is SRI Project FCI-6047.

This technical report has been reviewed and is approved.

*John T. Keke*

RAYMOND J. LAWRENCE, JR.  
Captain, USAF  
Project Officer

*Harry F. Rizzo*  
HARRY F. RIZZO  
Major, USAF  
Chief, Physics Branch

*Claude K. Stambaugh*  
CLAUDE K. STAMBAUGH  
Colonel, USAF  
Chief, Research Division

## ABSTRACT

## (Distribution Limitation Statement No. 2)

The pressure-volume-energy (P-V-E) equation of state of aluminum and Teflon has been investigated. The P-V-E equation of state of a material is needed to solve nonreactive flow problems using computer codes such as PUFF. Explosively induced shock waves have been used to generate Hugoniot curves for solid and porous specimens of aluminum and Teflon. For aluminum it is found that the following P-V-E equation of state can reproduce the experimental data quite nicely:

$$P = bE + g(V)$$

$b$  is a constant and  $g$  is an arbitrary function of  $V$ . This corresponds to  $\gamma$ ,  $N$  = constant, where  $\gamma$  is Gruneisen's ratio. Because of the spread of the Teflon data in the P-V plane, it was not possible to obtain a P-V-E fit. However, the data imply that Gruneisen's ratio varies from about 0.7 to 1.4 at high pressures. Gas gun experiments using manganin wire pressure recordings were performed. Sound speeds deduced from these records imply that Teflon becomes anomalous stiff under shock.

CONTENTS

<u>Section</u>	<u>Page</u>
I	1
INTRODUCTION	
II	3
EXPERIMENTAL TECHNIQUE	
1. Explosive Technique	3
2. Gas Gun Technique	3
3. Specimen Materials	3
III	5
EXPERIMENTAL RESULTS AND ANALYSIS	
1. Aluminum	9
2. Teflon	14
IV	29
CONCLUSIONS AND SUMMARY	
Appendix	33
References	35
Distribution	36

This page intentionally left blank.

## ILLUSTRATIONS

Figure	Page
1 Cross-Section View of Target Assembly Mounted on Nuzzle of Gas Gun	4
2 Target Assembly	6
3 Shock Velocity vs. Particle Velocity for Porous Aluminum	11
4 Hugoniot for Aluminum	12
5 Shock Velocity vs. Particle Velocity for Teflon	16
6 Teflon Hugoniot Data	17
7 Gage Record and Wave Profile for Shot 12,872--Second Gage	20
8 Gage Record and Wave Profile for Shot 12,897--	21
(a) First Gage	21
(b) Second Gage	22
9 Gage Record and Wave Profile for Shot 12,957--	24
(a) First Gage	24
(b) Second Gage	25
10 Gage Record and Wave Profile for Shot 12,873--	26
(a) First Gage	26
(b) Second Gage	27

## TABLES

Table	Page
I Solid and Porous Aluminum Hugoniot Data	10
II Teflon Hugoniot Data from Explosive Tests	15
III Teflon Hugoniot Data from Gun Shots	18
IV Teflon Sound Speed Data	30

SECTION I  
INTRODUCTION

In order to predict the shock wave propagation and flow resulting from a stress suddenly applied on the surface of a material, it is necessary to know the pressure-volume-energy (P-V-E) equation of state of the material under consideration. Equations of state for use in the PUFF code are of particular interest. It has been the practice to use a Mie-Gruneisen form of equation of state to describe solids. This equation of state is derived by using an assembly of harmonic oscillators as a model to describe solids. It is assumed that the logarithmic change of frequency with volume is the same for all frequencies and this leads to Gruneisen's ratio, which depends upon specific volume (Ref. 1). The purpose of this work has been to study the range of pressures and volumes for which it is valid to apply the Mie-Gruneisen equation of state to solids and to determine Gruneisen's ratio.

The work reported here is a continuation of work described in Reference 2. Specifically, the work has been aimed at obtaining equation of state data on aluminum and Teflon,\* using shock wave compression of porous specimens. The emphasis in the earlier work was on aluminum. The present work has concentrated on generating more Teflon data although some porous aluminum data were obtained to supplement the earlier data. The new aluminum data slightly shift one of the Hugoniot's that was obtained earlier.

The new high-pressure Teflon data are in good agreement with those obtained earlier. In addition to the high pressure data from explosive tests, some lower pressure data from gas gun experiments are presented. The low pressure data include manganin wire gage measurements of stress profiles at different depths in the target material.

\* Trademark, E. I. du Pont de Nemours and Co.

SECTION II  
EXPERIMENTAL TECHNIQUE

1. Explosive Technique

The explosive technique used to generate the high pressure aluminum and Teflon data is described in detail in Ref. 2. The technique consists of explosively accelerating a flat metal plate that impacts a 2024 aluminum driver plate upon which the samples are mounted. Measurements are made of the free-surface velocity of the aluminum driver plate and of the shock velocity through the sample. These data, combined with the loading and unloading curves in the pressure-particle velocity plane for the 2024 aluminum, are sufficient to compute Hugoniot points for the sample material using the impedance match technique (Ref. 3). In this technique, it is assumed that a steady state plane shock wave exists in the sample material and in the 2024 aluminum driver plate. Experiments on samples of varying thickness indicate that this assumption is valid for homogeneous samples if care is taken to prevent side and rear rarefactions from overtaking and interacting with the shock. The assumption is more questionable in very heterogeneous media such as granular or porous materials. The data have been treated here assuming steady state.

All measurements of shock and free-surface velocity were made optically by using a rotating mirror streak camera. The present data were obtained on 70-mm film using a Beckman and Whitley Model 770 camera. This camera has a maximum writing speed of 10 mm/usec as compared to a maximum of 3.84 mm/usec for the 35-mm-camera used in the earlier work.

2. Gas Gun Technique

Low-pressure experiments on Teflon were performed using a 2-1/2-inch-diameter gas gun projectile to generate shock waves. The projectile head is made either of Teflon to match the target or of 2024 aluminum. A schematic diagram of the shot assembly mounted at the muzzle of the gun is shown in Fig. 1. The projectile is accelerated down the evacuated

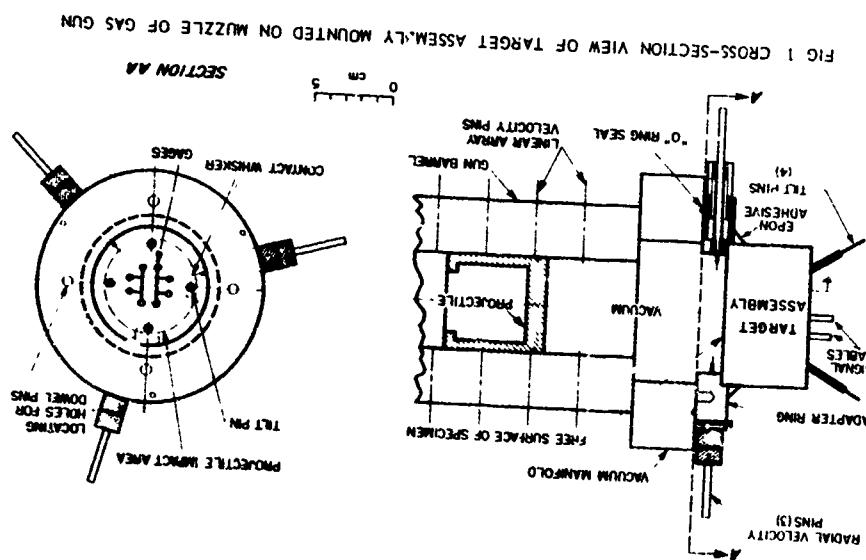
gun barrel by helium from a high pressure reservoir. The projectile velocity is measured near the muzzle of the barrel by a linear array of four pins spaced at intervals of one-tenth foot. The arrival of the projectile is also detected by three radial velocity pins placed in front of the specimen impact surface. Four tilt pins, passing through the target in a square array at 90° intervals, record the actual impact and provide a measure of the nonsimultaneity of impact (tilt). Tilt is minimized by carefully preparing flat target and projectile faces and aligning the target face perpendicular to the axis of the gun barrel. The target alignment technique has been described by Linde and Schmidt (Ref. 3).

Stress-time profiles in the target were recorded using manganin wire gages. Two gages were mounted in each target assembly as shown in Fig. 2. One gage wire is mounted on the impact surface and is covered by either a 0.001-inch layer of Mylar or a thin Teflon sheet. The second gage wire is placed between two Teflon disks, which are bonded together with a thin layer of epoxy. In making a stress measurement, a constant current is passed from a triggered power supply through the piezoresistive element from terminal A to A' (Fig. 2). The initial resistance and the change in resistance of the element due to passage of the wave are measured by recording the voltage B-B' as a function of time. The ratio of the change in resistance to the initial resistance (this is equivalent to the ratio of the change in voltage to the initial voltage for a constant current) is proportional to the stress. A detailed description of the operation of the gage is given in Ref. 4.

### 3. Specimen Materials

The porous aluminum specimens were cold pressed to the desired density from Reynolds No. 40 atomized aluminum powder, which is greater than 99 percent pure aluminum. For specimens of porosity  $m = 1.4$  and  $\rho = 1.7$ ,\*

\* Porosity  $m$  is defined by  $m = \rho_0 / \rho_{app}$  where  $\rho_0$  is the crystal density at standard conditions and  $\rho_{app}$  is the apparent density under the same conditions.



the powder was screened and only those particles less than 200  $\mu$  were used. The specimens were quite strong mechanically, machinable, and easily handled. Specimens of porosity  $m = 2.0$  were made of powder between 200  $\mu$  and 300  $\mu$ . These were somewhat more fragile. The raw material and specimen preparation were reported in Ref. 2. Specimens were nominally 1/8 inch thick by 1 inch in diameter.

The solid and porous Teflon were obtained from the Avco Corporation, where low pressure work on aluminum and Teflon was also performed. The preparation of these samples is described in Ref. 5.

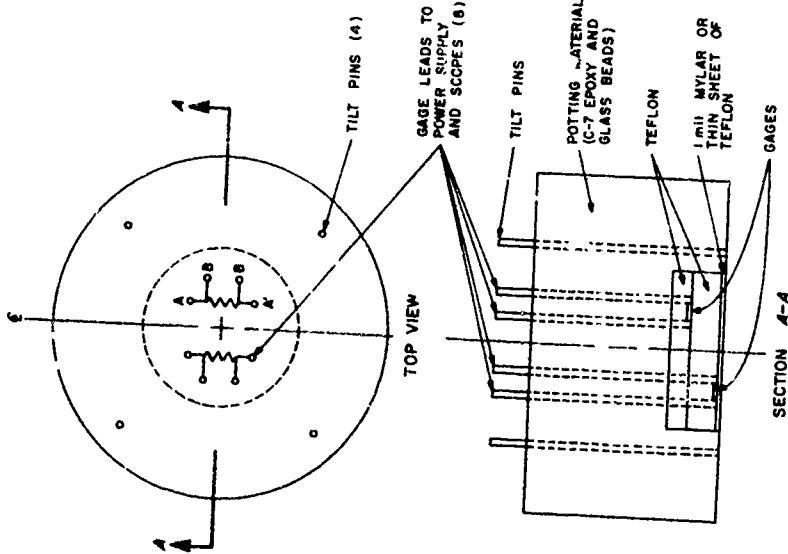


FIG 2 TARGET ASSEMBLY

SECTION III  
EXPERIMENTAL RESULTS AND ANALYSIS

1. Aluminum

The Hugoniot data for solid and porous aluminum are presented in Table I. All of the data reported in Ref. 2 are included so that Table I is a complete summary. The data for porous aluminum are plotted in the shock velocity-particle velocity plane in Fig. 3. The straight lines in the figure are drawn as guides and do not represent fits to the data. The Hugoniot in the pressure-to-ume plane are presented in Fig. 4 (the labels  $t = 3/16$  in. and  $t = 3/8$  in. refer to the sample thickness). The Hugoniot for solid aluminum was taken from the work of Al'tsuler et al. (Ref. 6). They quote a value of 2.71 g/cm<sup>3</sup> for the initial density of aluminum. To make their Hugoniot comparable to the data reported here on aluminum of initial density 2.70 g/cm<sup>3</sup>, each of their volumes at a given pressure was changed by a factor of 1.71270. This maintains the shape of the curve but translates it slightly to the right, i.e., toward larger specific volumes. The solid Hugoniot curve was then fitted by

$$P_H = 16899.5 - 13437.5V + 363023V^2 - 333405V^3 \quad (1)$$

where  $V$  is specific volume in cm<sup>3</sup>/g and  $P_H$  is Hugoniot pressure in kilobars. To fit the porous aluminum pressure-volume (P-V) curve, smooth curves were drawn through the points. From these curves and the solid aluminum Hugoniot, the internal energy as a function of pressure at a given volume was computed by using the Hugoniot energy relation

$$E_H = E_0 + (1/2)P_H(mV_0 - V) \quad (2)$$

where  $m$  is the porosity. The values of the initial internal energy  $E_0$  and the initial specific volume  $V_0$  were taken as  $1.66 \times 10^5$  ergs/g and

Table 1  
SOLID AND POROUS ALUMINUM HUGONIOT DATA

Shot No.	Initial Volume (cm <sup>3</sup> /gm)	Shock Velocity (mm/sec)	Particle Velocity (mm/sec)	Pressure (kbar)	Fluid Volume (cm <sup>3</sup> /gm)	Internal Energy Change (10 <sup>9</sup> erg/gm)
$m = 1.0$						
11,477	0.370	7.31	1.45	286	0.297	10.65
11,370	0.370	7.75	1.80	377	0.284	16.21
11,563	0.370	8.77	2.46	583	0.266	30.31
11,460	0.370	9.06	2.58	633	0.263	33.23
11,562	0.370	9.13	2.71	667	0.260	36.36
$m = 1.4$						
11,477	0.519	5.08	1.88	194	0.328	17.57
12,778	0.518	5.17	1.89	189	0.328	17.95
12,791	0.518	5.36	1.84	191	0.339	17.09
12,283	0.518	6.43	2.45	304	0.321	30.09
12,535	0.518	6.83	2.70	357	0.313	36.33
12,908	0.518	6.90	2.77	368	0.310	38.27
10,591	0.520	7.13	3.03	418	0.299	46.18
10,552	0.521	7.79	3.42	511	0.293	58.25
12,536	0.521	7.97	3.45	528	0.286	59.40
$m = 1.7$						
11,155	0.629	5.43	2.43	210	0.347	29.61
11,286	0.631	5.82	2.67	246	0.341	35.67
11,682	0.629	5.93	2.69	254	0.344	36.19
10,878-1	0.630	5.58	3.05	319	0.338	46.57
10,878-2	0.629	6.50	3.06	316	0.333	46.76
10,926	0.631	6.83	3.20	347	0.335	51.35
11,330	0.631	6.90	3.25	355	0.334	52.72
10,581	0.633	6.91	3.26	356	0.334	53.22
10,894	0.637	7.29	3.44	397	0.333	58.95
11,305	0.631	7.49	3.58	426	0.330	64.11
10,592	0.645	7.55	3.66	496	0.327	67.14
$m = 2.0$						
11,153	0.741	5.09	2.58	177	0.365	33.19
11,286	0.741	5.57	2.82	214	0.364	33.85
11,762	0.741	5.64	2.85	218	0.367	40.16
10,926	0.741	5.72	3.37	305	0.369	56.73
11,330	0.740	6.79	3.41	313	0.369	58.06
10,894	0.741	7.16	3.60	348	0.363	64.90
11,379	0.744	7.29	3.85	377	0.351	74.08
11,305	0.741	7.32	3.78	373	0.360	71.05

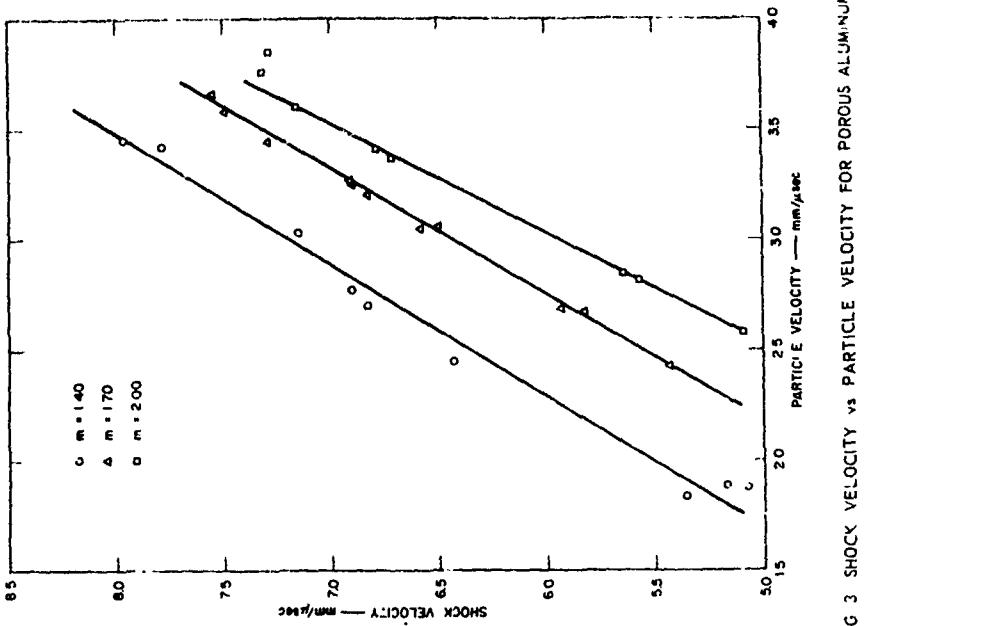


FIG 3 SHOCK VELOCITY vs PARTICLE VELOCITY FOR POROUS ALUMINUM

0.370  $\text{cm}^3/\text{gm}$ , respectively. As can be seen from Fig. 4, a maximum of three energy-pressure points can be computed at a given volume in the volume range where the Hugoniots for  $m = 1$ , 1.4, and 1.7 overlap.

Several points can be computed near  $V = V_0$  from the Hugoniots for  $m = 2$ . Plots of the energy vs. pressure at constant volume are straight lines, indicating that  $(\partial P/\partial E)_V$  is a function only of volume; furthermore, the fact that all slopes are nearly the same indicates that  $(\partial P/\partial E)_V$  is nearly constant. The energy-pressure-volume equation of state data can then be fitted by a form

$$P = bE + g(V) \quad (3)$$

where  $b = (\partial P/\partial E)_V$ , a constant, and  $g$  is an arbitrary function of  $V$ . From the pressure-energy plots, an average value of  $b = 5.15 \text{ gm/cm}^3$  was obtained. The function  $g(V)$  was computed by substituting Eqs. (1) and (2) into Eq. (3) for  $P$  and  $E$ , respectively. A fit for  $g(V)$  was then found to be given by

$$g(V) = 8721.138 - 66012.96V + 171668.7V^2 - 154117.1V^3 \quad (4)$$

where  $V$  is in  $\text{cm}^3/\text{gm}$  and  $g(V)$  is in kilobars. The Hugoniots curves drawn in Fig. 4 were generated from the equation of state given by Eq. (3) with  $b = 5.15 \text{ gm/cm}^3$  and  $g(V)$  given by Eq. (4). This form of the equation of state is a Mie-Gruneisen equation with  $\gamma/V$  a constant, where  $\gamma$  is Gruneisen's ratio.

The Hugoniots curves in the pressure-volume plane for the various porosities are given by

$$P_H(V) = \frac{bE_0 + g(V)}{1 - \frac{b}{2}(mV_0 - V)} \quad (5)$$

The minimum volume to which a porous material can be compressed is given by

$$V_{\min} = mV_0 - \frac{2}{b} \quad (6)$$

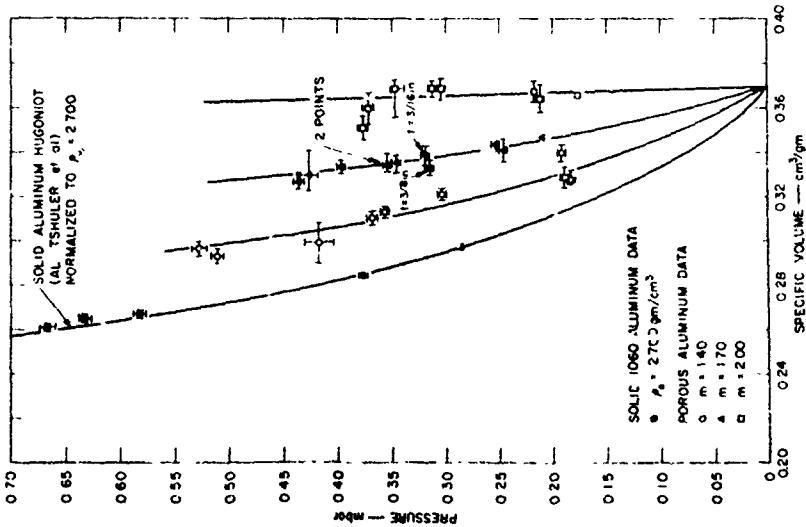


FIG. 4 HUGONIOTS FOR ALUMINUM

Setting  $V_{min} = V_o$  in Eq. (6) shows that the Hugoniot of aluminum of porosity  $m = 2.05$  would be a vertical line passing through  $V_o = 0.370 \text{ cm}^3/\text{gr}$ . The Hugoniot curves for  $m = 1.4$ , 1.7, and 2.0 drawn in Fig. 4 were generated by Eq. (5). With the exception of the highest pressure point for  $m = 2.0$ , Eq. (5) seems to fit the data quite well.

A fit to the data using  $m = (r = c/c_o - 1)$  as a variable is discussed in the Appendix.

## 2. Teflon

The Hugoniot data for Teflon obtained from the explosive tests are presented in Table II. Plots of the data in the shock velocity-particle velocity plane and the pressure-volume plane are presented in Figs. 5 and 6, respectively. The four lowest pressure points for solid Teflon on Figs. 5 and 6 are from gas gun shots (Table III). These four points and the lowest pressure explosive data point for solid Teflon exhibit a slightly steeper slope in the shock velocity-particle velocity plane than do the remaining points. This change in slope produces the cusp shown by the dashed line in the pressure-volume diagram. However, the lowest pressure explosive data point is questionable because the optical record was poor, and so the cusp is shown as a dashed line to indicate that not too much significance should be attached to it.

Since the Hugoniot data for the two porosities of Teflon ( $m = 1.42$  and  $m = 2.77$ ) do not overlap in volumes, only two energy-pressure points can be computed for any one volume. The uncertainties in the volume at a given pressure on the Hugoniots of porous Teflon are also quite large.

Values of the average Gruneisen ratio as calculated from the offsets of the Hugoniots in Fig. 6 run from 1.2 to 1.4 when calculated between the solid and  $m = 2.77$  Hugoniots. Quite large internal energy increases occur along the Hugoniot of porosity  $m = 2.77$ , and it is quite possible that some decomposition takes place.

The gage records for gas gun shots (Table III) generally indicate a lower pressure than that computed by the impedance match technique using the projectile velocity and the shock velocity in the sample.

Table II  
TEFLON HUGONIOT DATA FROM EXPLOSIVE TESTS

Shot No.	Initial Density (gm/cm <sup>3</sup> )	Shock Velocity (mm/usec)	Particle Velocity (mm/usec)	Pressure (kbar)	Final Volume (cm <sup>3</sup> /gm)	Energy Increase (10 <sup>5</sup> erg/gm)	
						m = 1.00	m = 1.42
12,495	2.157	3.529	1.075	82.0	0.3224	5.73	
12,778	2.168	5.120	1.833	203.0	0.2959	16.78	
11,570	2.169	5.589	2.215	289.0	0.2782	24.58	
12,284	2.175	5.628	2.205	269.5	0.2760	24.75	
12,283	2.170	5.994	2.415	314.0	0.2752	29.56	
12,535	2.169	6.360	2.665	368.0	0.2678	35.54	
11,563	2.163	6.830	2.945	435.0	0.2629	43.37	
11,460	2.172	7.105	3.080	476.0	0.2608	47.50	
11,562	2.163	7.262	3.200	504.0	0.2582	51.35	
12,536	2.169	7.543	3.375	552.5	0.2547	57.26	
12,495	1.533	2.639	1.255	50.5	0.3442	7.77	
12,778	1.524	4.354	2.120	140.5	0.3366	22.45	
11,570	1.517	4.906	2.555	190.3	0.3160	32.64	
12,284	1.525	4.952	2.535	191.5	0.3201	32.13	
12,535	1.516	5.774	3.065	268.0	0.3094	46.92	
11,563	1.524	6.207	3.373	319.0	0.2995	56.87	
11,562	1.515	6.683	3.680	371.0	0.2981	66.97	
12,536	1.516	6.992	3.855	408.5	0.2959	74.28	
12,495	0.7967	2.048	1.420	23.0	0.3849	10.00	
12,778	0.7606	3.741	2.493	71.0	0.4385	31.10	
11,570	0.7828	4.374	2.985	103.0	0.4057	44.87	
12,284	0.7782	4.340	2.980	100.8	0.4027	44.46	
12,535	0.7870	5.129	3.600	146.0	0.3788	65.13	
11,563	0.7782	5.646	3.985	175.5	0.3780	79.58	
11,562	0.7828	6.086	4.335	207.0	0.3676	84.12	
12,536	0.7869	6.419	4.565	231.0	0.3670	104.4	

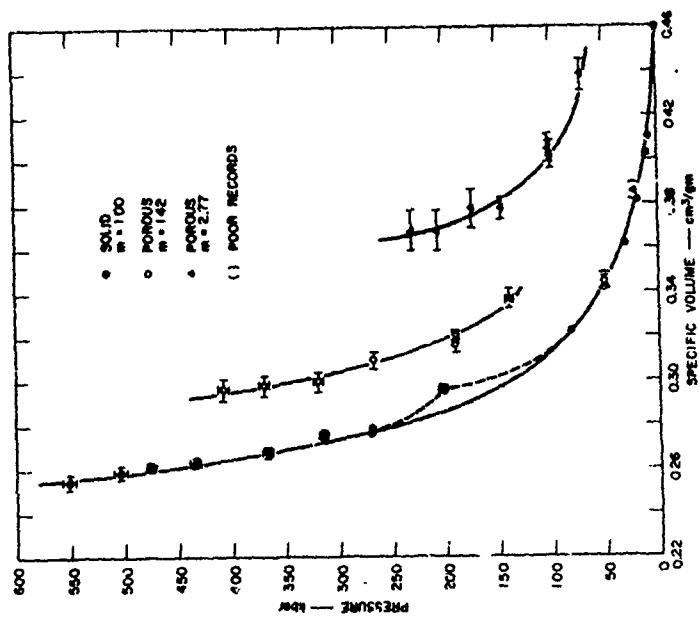


FIG. 6 TEFILON HUGONIOT DATA

17

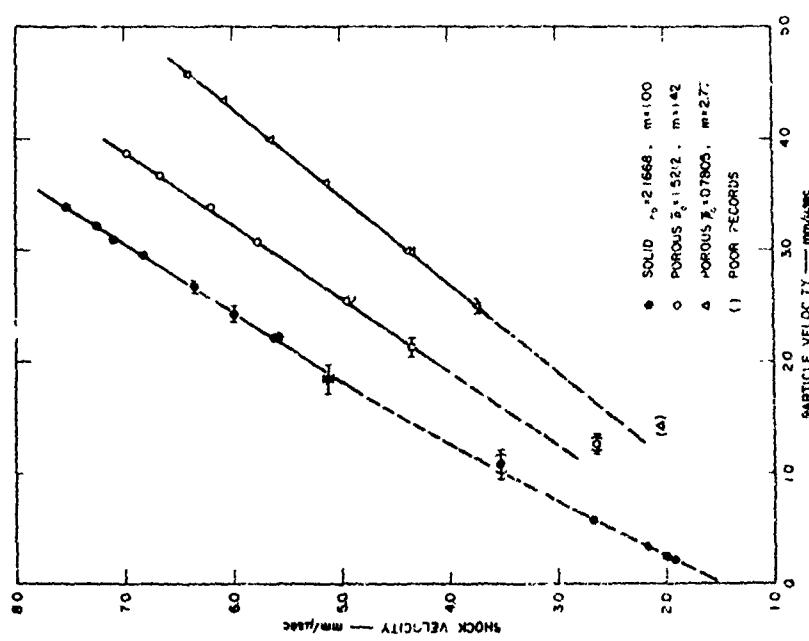


FIG. 5 SHOCK VELOCITY vs. PARTICLE VELOCITY FOR TEFILON

16

In Shot 12,872 the projectile head was a 3.03 mm disk of Teflon backed by aluminum. The Teflon target had one gage on the impact surface covered by a 0.001-inch layer of Mylar and a second gage 6.44 mm from the impact surface. Due to a timing error, the fast sweep oscilloscope record from the first gage was lost. A plot of the record from the second gage is shown in Fig. 7. This gage record indicates a double wave structure with a 2 to 3 kbar wave leading the 8 kbar peak wave by about 0.15  $\mu$ sec. This structure was also observed on the long duration records of the second gage but not on the long duration record of the first gage, and thus exhibits the properties of a propagating double wave. Upon impact a shock is induced in the Teflon projectile head, which propagates back and reflects from the aluminum projectile as a second shock into the Teflon. The arrival of this second shock at the deeper gage produces the second pressure rise seen at about 2.25  $\mu$ sec after arrival of the first wave. The final stress behind this wave is 12.3 kbar and a distance-time construction of the wave and particle paths indicates that the velocity of the reflected shock is about 3 mm/ $\mu$ sec with respect to the material.

For Shot 12,897 the projectile head was of 2024 aluminum. The peak pressures recorded by the two gauges were somewhat higher than in Shot 12,872. The wave profiles, shown in Fig. 8, do not exhibit the "precursor" structure that was seen in the preceding shot (12,872). Both gage records show a very rapid pressure rise to 7 or 8 kbar and then a more gradually rounded rise to the peak stress. Similar behavior has been observed in polyethylene (Ref. 7). The decrease in pressure late in the gage record is due to the arrival of the rarefaction originating at the rear surface of the projectile. The time of arrival of this rarefaction indicates a sound speed of 2.82 mm/ $\mu$ sec at about 9 kbar in Teflon.

19

<sup>a</sup>Gage 2 is approximately 6.4 mm behind and Gage 1 except shot 12,897, in which it is 4.76 mm from Gage 1.

<sup>b</sup>From shock and projectile velocities.  
<sup>c</sup>Gage on impact surface covered by 1 mil Mylar.  
<sup>d</sup>Gage 1.5 mm from impact surface.  
<sup>e</sup>From impedance match using measured shock velocity.

<sup>f</sup>Gage 1.7 mm from impact surface.

Shot No.	Projectile		Shock		Impedance		Match		Gage 1		Gage 2 <sup>a</sup>	
	Velocity (mm/ $\mu$ sec)											
12,872	0.417	Teflon	1.92	8.7	8.6 <sup>c</sup>	8.6 <sup>c</sup>	8.1	8.1	0.208	0.410	0.247 <sup>d</sup>	0.247 <sup>d</sup>
12,897	0.417	Teflon	1.92	8.7	8.6 <sup>c</sup>	8.6 <sup>c</sup>	8.1	8.1	0.208	0.410	0.247 <sup>d</sup>	0.247 <sup>d</sup>
12,857	0.311	Aluminum	2.00	10.8	10.8	10.8	9.1	9.1	0.247 <sup>d</sup>	0.403	0.370	0.370
12,873	0.781	Aluminum	2.67	17.6	17.6	17.6	15.6	15.6	0.370	0.392	0.528 <sup>e</sup>	0.528 <sup>e</sup>

Table III

FIG. 8 GAGE RECORD AND WAVE PROFILE FOR SHOT 12,897

(a) FIRST GAGE

21

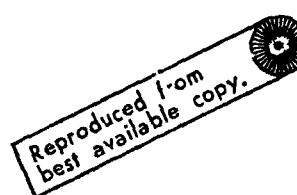
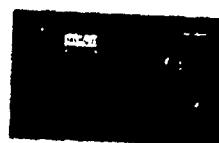


FIG. 7 GAGE RECORD AND WAVE PROFILE FOR SHOT 12,872 SECOND GAGE

22

SHOT 12,872 WAVE PROFILE  
REPRODUCED FROM MICROFILM



Shot 12,857 had a Teflon projectile head, which was 3.12 mm thick, backed by 2024 aluminum. The Teflon target had one gage on the surface covered by 0.001 inch Mylar and a second gage imbedded in Teflon 6.47 mm from the impact surface. The oscilloscope records and wave shapes from these records are plotted in Fig. 9. A polarization signal, possibly from shocking the Mylar cover, is observed before the pressure rise on the record from the first gage. Both gages record some oscillations before settling down to record the final peak pressure. However, the overshoot of pressure indicated by the first gage is not recorded by the second gage. Again the arrival of the reflected shock originating at the interface between Teflon and aluminum on the projectile is seen to arrive at the first gage about 2.25  $\mu$ sec after the first wave and at the second gage at about 0.8  $\mu$ sec after the arrival of the first wave. Recall that the time bases on the gage records begin with the arrival of the first wave at the gage position.

In the highest pressure gun shot, 12,872, an aluminum projectile struck the Teflon target. The oscilloscope records and plots of the wave shapes are presented in Fig. 10. The records indicate a sounding of the wave front and a decrease in peak pressure amplitude as the wave propagates from gage 1 to gage 2. The small oscillation that appears on the wave front was observed in varying degrees on all but the lowest pressure shot. Both gage records show a decrease in pressure resulting from the arrival of the rarefaction originating from the back surface of the projectile. From the time of arrival of this rarefaction a sound speed of 3.77  $\text{mm}/\mu\text{sec}$  at 34 kbar was computed. Note in the record from gage 2 the arrival of a shock reflected from the potting compound behind the Teflon target.

FIG. 6 (Concluded)

(b) SECOND GAGE

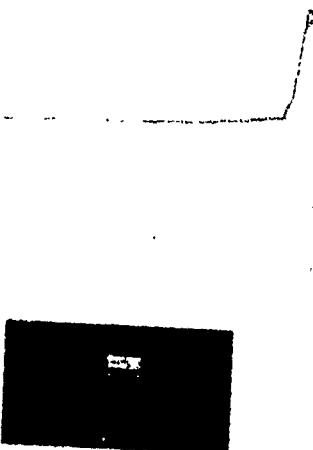
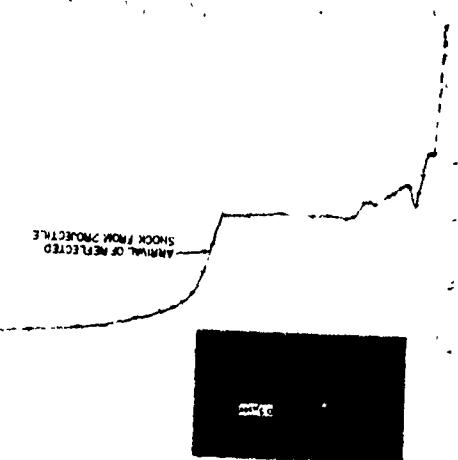


FIG. 9 (Concluded)

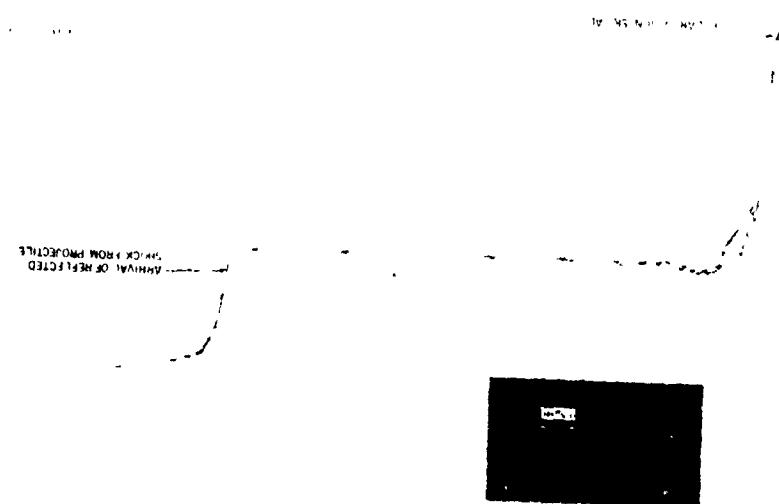
(b) SECOND GAGE



25

FIG. 9 GAGE RECORD AND WAVE PROFILE FOR SHOT 12,857

(a) FIRST GAGE



24

FIG 10 (Concluded)

(b) SECOND GAGE

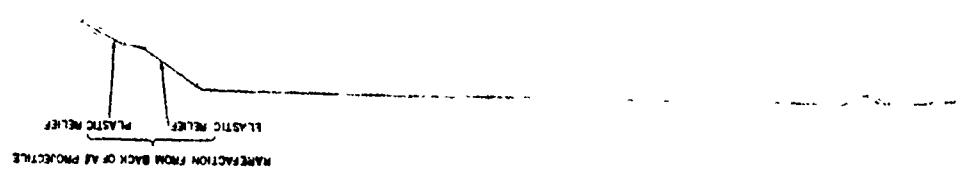
27



FIG 10 GAGE RECORD AND WAVE PROFILE FOR SHOT 12,873

(a) FIRST GAGE

26



#### SECTION IV

##### CONCLUSIONS AND SUMMARY

The pressure-volume-energy data for aluminum in the region of the P-V plane bounded by the Hugoniot of solid aluminum to 700 kbar and the vertical line through  $V_0 = 0.370 \text{ cm}^3/\text{gm}$  up to about 400 kbar are well represented by a simple Mie-Gruneisen equation of state of the form

$$P = bE + g(V) \quad (3a)$$

This page intentionally left blank.

$$g(V) = 8721.138 - 66012.96V + 171688.7V^2 - 154117.1V^3 \quad (4a)$$

This form of the Mie-Gruneisen equation in which the pressure is directly proportional to the internal energy at a given volume implies that Gruneisen's ratio  $\frac{1}{\Gamma} = V(\frac{\partial P}{\partial E})_V$  is proportional to volume.

Owing to the steepness of the Hugoniot curves in the P-V plane it is extremely difficult to accurately determine an average Gruneisen's ratio from offsets in pressure and energy at a given volume. The shock wave data give the locus of points on the P-V-E surface but, because of the steepness of the surface compared with the precision of the experiments, it is very difficult to obtain accurate values for the derivatives of the surface. What has been done by introducing Eq. (3) has been to assume a form of P-V-E equation of state and force it to fit the data. There may be other analytical forms that fit the data equally well or perhaps even better. However, the simplicity of Eq. (3) in which pressure is proportional to energy would seem to make it desirable for computational work.

There is not enough overlap in volume of the Hugoniot's of solid and porous Teflon to permit more than two energy-pressure points to be obtained at one volume. Values of Gruneisen's ratio computed from the offsets in pressure and energy along the Hugoniot curves vary by a factor of 2 (0.7 to 1.4). The zero pressure value estimated from static values of the bulk modulus, thermal expansion coefficient, and specific heat is about 0.313 (Ref. 2).

Sound speeds computed from the time of arrival at the gages of the rarefaction originating from the back surface of the aluminum projectiles on Shots 12,897 and 12,873 appear anomalously high. An estimate of the slope of the Hugoniot in the P-V plane can be obtained from the slope of the Teflon Hugoniot in the shock velocity-particle velocity plane (U-u plane). Since the U-u Hugoniot has less curvature than the P-V Hugoniot, the derivative  $du/du$  can probably be more accurately determined from a plot of the data than can the derivative  $dp/dv$ . The two derivatives are related by

$$\frac{dp}{dv} = c_o^2 u^2 \left[ \frac{u \frac{du}{du} + u}{u \frac{du}{du} - u} \right] \quad (7)$$

The "Hugoniot sound speed"  $c_H$  is then defined by

$$c_H^2 = -u^2 \frac{dp}{dv} \quad (8)$$

$c_H$  should be an upper limit for the sound speed of a hydrodynamic material. From a graphical evaluation of the slope of the U-u curve for Teflon at the Hugoniot points for Shots 12,897 and 12,873, values of  $c_H$  have been estimated (Table IV).

Table IV

TEFLON SOUND SPEED DATA					
Shot No	Pressure (kbar)	U (mm/ $\mu$ sec)	u (mm/ $\mu$ sec)	$\frac{du}{du}$	$c_H$ (mm/ $\mu$ sec)
12,897	10.8	2.00	0.208	2.22	2.37
12,873	33.5	2.67	0.578	2.00	3.31

APPENDIX  
ANALYTICAL FITS TO THE EXPERIMENTAL DATA

The cubic fit to the Rugoniot of solid aluminum is given by Eq. (1). The coefficients are quite large numbers of oscillating sign, so that the pressure at a given volume is given by a sum of small differences of large numbers. This behavior indicates that a cubic fit to the data is probably not the best form. However, time does not permit further investigation of fits to the data to be included in this report.

For use in PUPP the variable  $\mu = \rho/\rho_0 - 1$  is used rather than the specific volume. A fit to the Rugoniot of solid aluminum in the variable  $\mu$  is

$$P_H = 778.518\mu + 1165.95\mu^2 + 1416.69\mu^3 \quad (9)$$

where  $P_H$  is in kilobars. Here all coefficients are of the same sign. In the same manner in which the arbitrary function  $g(V)$  was determined, an arbitrary function  $F(\mu)$  was found to fit the equation of state to the form

$$P(\mu, E) = bE + F(\mu) \quad (10)$$

the function  $F(\mu)$  is given by

$$F(L) = -8.375 + 768.28\mu + 545.5\mu^2 + 523.97\mu^3 \quad (11)$$

Another form in which Eq. (10) can be used is

$$P(\mu, E) = P_H(L) \left[ 1 - \frac{b\mu}{20} \right] + b(E - E_0) \quad (12)$$

where  $P_H(u)$  is given by Eq. (9) and  $b = 5.15 \text{ gm/cm}^3$ . Comparing Eqs. (10) and (12) it is seen that  $F(u)$  is simply a fit to

$$F(u) = P_H(u) \left[ 1 - \frac{bu}{2c} \right] - bE_0 \quad (13)$$

#### REFERENCES

1. Rice, N. H., R. G. McQueen, and J. M. Walsh, "Compression of Solids by Strong Shock Waves," Solid State Physics, F. Seitz and D. Turnbull, eds., Academic Press, New York, 1958, Vol. 6.
2. Anderson, G. D., A. L. Fahrebruch, and D. G. Doran, "Equation of State of Solids, Aluminum and Teflon," AFWL-TR-66-147, August 1965.
3. Linde, R. K., and D. N. Schmidt, "Measuring the Sub-microsecond Response of Shock-Loaded Materials," Rev. Sci. Inst. 37, 1 (1966).
4. Keough, D. D., and W. Wilkinson, "Manganin-Wire Shock-Transducer Investigation," AFWL-TR-65-170, December 1965.
5. Morgan, D. T., M. Rockowitz, and A. L. Atkinson, "Measurement of the Gruneisen Parameter and the Internal Energy Dependence of the Solid Equation of State of Aluminum and Teflon," AFRL-TR-65-117, October 1965.
6. Al'tshuler, L. V., S. B. Kormer, A. A. Bakunova, and R. F. Trunin, "Equation of State for Aluminum, Copper, and Lead in the High Pressure Region," Soviet Phys./JETP II, 373 (1960).
7. Anderson, G. D., W. J. Murri, R. C. Alversen, and S. V. Banaguid, "Stress Relaxation in the Shock Compression of Solids, Final Report SRI Project FG-5783, Contract AF 29(601)-719, AFWL-TR-67-24, February 1967.

## DISTRIBUTION

## DISTRIBUTION (cont'd)

## No. CUS

HEADQUARTERS USAF

1 Hq USAF (AFTAC/TD-3, Capt Herman), Wash, DC 20330

1 USAF Dep, The Inspector General (AFTIC), Norton AFB, Calif 92409

1 USAF Directorate of Nuclear Safety (AFINS), Kirtland AFB, NM 87117

MAJOR AIR COMMANDS

1 AFSC, Andrews AFB, Wash, DC 20331

1 (SCISW)

1 AUL, Maxwell AFB, Ala 36112

1 USAFIT-L, Wright-Patterson AFB, Ohio 45433

AFS ORGANIZATIONS

1 AFSC STLO, R&T Div, AFUPO, Los Angeles, Calif 90045

1 AFSC STLO (RFSAS), Suite 104, 363 S. Taaffe Av, Sunnyvale, Calif 94086

1 FTD (TDBT), Wright-Patterson AFB, Ohio 45433

5 AF Materials Laboratory, Wright-Patterson AFB, Ohio 45433

1 AF Avionics Laboratory, Wright-Patterson AFB, Ohio 45433

1 AF Flight Dynamics Laboratory, Wright-Patterson AFB, Ohio 45433

1 AF Aero-Propulsion Laboratory, Wright-Patterson AFB, Ohio 45433

1 ASD, Wright-Patterson AFB, Ohio 45433

(ASAPR)

1 (ASRCP)

1 (ASAMC)

SEG, Wright-Patterson AFB, Ohio 45433

(SEPIR)

2 (SESOS, Mr. J. L. Wilkinson)

1 ORA (RRD), Holloman AFB, NM 88330

1 AEDC (AEYD), Arnold AFB, Tenn 37289

1 FSD (ESTI), L. G. Rapicom Flld, Bedford, Mass 01731

2 AFPG (PGBPS-12), Eglin AFB, Fla 32542

1 RADC (EMHAL-1), Griffiss AFB, NY 13442

1 AFRL (RPT), Edwards AFB, Calif 93323

SAMS, AFUPO, Los Angeles, Calif 90045

1 (SSSD)

1 (SSIM)

1 (SSIDS)

1 (SSSE)

1 (SMT)

1 (SMY)

1 (SMSE, Capt R. Jackson)

1 (SMTG)

1 (SMC)

1 (SMISH-1)

KIRTLAND AFB ORGANIZATIONS

AFSWC, Kirtland AFB, NM 87117

1 (SWER)

1 (SWT)

1 ADC (ADSHO, Special Weapons Office, Kirtland AFB, NM 87117)

1 AFRL, Albuquerque Liaison Office (ACMNOQ), Kirtland AFB, NM 87117

1 SAC Res Rep (SACLO), Kirtland AFB, NM 87117

1 TAC Liaison Office (TACLO-S), Kirtland AFB, NM 87117

AFML, Kirtland AFB, NM 87117

12 (NLIL)

2 (VLAA)

1 (WLDC)

3 (WLDN)

1 (WLK)

2 (WLRE, Dr. Guenther, Lt Assay)

15 (WLRE)

1 (WLK, Mr. Aubrey)

1 (LX)

1 (WLW)

## OTHER AIR FORCE AGENCIES

Director, USAF Project RAND, via: AFLO, The RAND Corp, 1700 Main St, Santa Monica, Calif 90406

1 (RAND Physics Div)

1 (RAND Library)

1 (Dr. Olin Nance)

## DISTRIBUTION (cont'd)

No. cys

1 OAR (ROS), 1400 Wilson Blvd, Arlington, Va 22209  
 1 AFSSR, 1400 Wilson Blvd, Arlington, Va 22209  
 1 AFCA, 1. G. Hanscom Fld, Bedford, Mass 01731  
 1 Office of Asst Secy AF, Asst for Laboratories, ATTN: Dr. W. L. Lehmann, Wash, DC 20301

**ARMY ACTIVITIES**

1 Chief of Research and Development, Dept of the Army (CRD/P, Scientific and Technical Info Div), Wash, DC 20310  
 1 Commanding Officer, Harry Diamond Laboratories, ATTN: Library, Wash, DC 20438  
 1 US Army Material Command, NIKE-X Fld Ofc (WMCPCM-NXE-FB), Ls Col F. G. Thomas, Bell Telephone Laboratories, Inc, Whippany, NJ 07981  
 1 Redstone Scientific Information Center, US Army Missile Command (Chief, Document Section), Redstone Arsenal, Ala 35801  
 1 Commanding Officer, Ballistic Research Laboratories, Aberdeen Proving Ground, Md 21015  
 1 (AMGR-TB, Mr. J. Mazzares)  
 1 (AMGR)  
 1 Commanding Officer, Picatinny Arsenal, Dover, NJ 07801  
 1 (ShAPA-UCI, Samuel Feltman Research Laboratories)

**(ShAPA-TN, Nike-X Engineering Division)**

1 Commanding Officer, US Army Research Office-Durham, Box CM, Duke Station, Durham, NC 27706  
 1 Chief of Engineers (ENGRC-EM), Dep of the Army, Wash, DC 20315  
 1 Director, Army Resch Ofc, 3045 Columbia Pike, Arlington, Va 22206  
 1 Director, US Army Engineer Research and Development Laboratories, ATTN: ERINFO Branch, Ft Belvoir, Va 20260  
 1 Commanding General, White Sands Missile Range (Tech Lab), White Sands, NM 87502

**NAVY ACTIVITIES**

1 Office of the Chief of Naval Operations, Dept. of the Navy, Wash, DC 20330  
 1 (OP-75, ATTN: D.R. Economic Energy Div)  
 1 (OP-754)  
 1 Chief of Naval Research, Dept of the Navy, Wash, DC 20390  
 1 Commanding Officer, Naval Research Laboratory, Wash, DC 20390  
 2 Commanding Officer and Director, US Naval Radiological Defense Laboratory, San Francisco, Calif 94135

## DISTRIBUTION (cont'd)

No. cys

1 Commanding Officer and Director, Navy Electronics Laboratory (Code 4223), San Diego, Calif 92152  
 1 Commanding Officer and Director, Naval Ship Research and Development Center, Wash, DC 20007  
 1 Commanding Officer and Director, Naval Civil Engineering Laboratory, Port Huron, Calif 93041  
 1 Commanding Officer and Director, Naval Applied Science Laboratory, Brooklyn, NY 11251  
 1 Commander, Naval Ordnance Test Station (Code 753), China Lake, Calif 93357  
 1 Commander, Naval Ordnance Laboratory, ATTN: Dr. Pollin, White Oak, Silver Spring, Md 20310  
 1 Director, Special Projects Office, Dept of the Navy, Wash, DC 20360  
 1 Office of Naval Research, Wash, DC 20360  
 1 Director of Naval Warfare Analyses, Institute of Naval Studies, Ofc of Chief Navy ps, 565 Technology Sq, Cambridge, Mass 02239  
 1 Commanding Officer, NMETP (Code WE), Kirtland AFB, NM 87117  
 1 Ordnance Systems Command, Dept of the Navy, Wash, DC 20360  
 1 Director, Laboratory Programs, Naval Material Command, ATTN: Dr. G. N. Johnson, Wash, DC 20360

**OTHER DOD ACTIVITIES**

1 Director, DASA, Wash, DC 20390  
 (Document Library Branch)  
 2 Co "ander, Fld Cad, DASA, Sandia Base, NM 87115  
 (Maj Lowery, J. Moulin)  
 1 (FCAG3, Spec Wpsn-Pub Div)  
 2 (Col Neutr, Tech Comd)

**DDC (TIAA), Cameron Station, Alexandria, Va 22314**

1 Director, ARP, DOD, The Pentagon, Wash, DC 20301  
 1 Ofc Dir Defense Resch & Eng, ATTN: J. Jackson, Ofc Atomic Programs, Rm 3E1071, The Pentagon, Wash, DC 20330  
 1 Director, Weapons Systems Evaluation Group, Wash, DC 20305  
 1 Director, ARP, DOD, The Pentagon, Wash, DC 20301  
 1 Ofc Dir Defense Resch & Eng, ATTN: J. Jackson, Ofc Atomic Programs, Rm 3E1071, The Pentagon, Wash, DC 20330  
 1 DDC (TIAA), Cameron Station, Alexandria, Va 22314

**AEC ACTIVITIES**

1 USAEC (HQ Lib, Rpts Sect), Mail Sta G-017, Wash, DC 205  
 1 USAEC (Chief, Div Tech Info Ext), Box 62, Oak Ridge, Tenn. 37831  
 1 ORNL (Tech Info Div), Berkeley, Calif 94720  
 1 Brookhaven National Laboratory, Upton, Long Island, NY 11973  
 1 Oak Ridge National Lab (Tech Lib), Oak Ridge, Tenn 37831

## DISTRIBUTION (cont'd)

No. cys

1 Sandia Corp, Box 5800, Sandia Base, NM 87115  
(C. D. Broyles)  
(C. D. Lundergan)  
(L. M. Barker)  
(W. Herrmann)  
(B. N. Butcher)  
(W. B. Benedict)  
(R. Graham)  
(R. Clem)  
(T. B. Lane)  
(Information Distribution Division)  
Sandia Corp, P. O. Box 969, Livermore, Calif 94551  
(Tech Library)  
(P. Glidde)  
(A. M. Blackwell)  
UCLR, ATTN: Director's Ofc, P. O. Box 868, Livermore, Calif 94551  
(Dr. W. G. Magnuson, Jr.)  
(R. Duff)  
(C. J. Taylor, D. Gamble)  
(Dr. R. B. Carr, Mech Eng Dept, L-722)  
Director, LASI, P. O. Box 1663, Los Alamos, NM 87554  
(Helen Redman, Rpt Lib)  
(D. Myers)  
(W. Deal)  
(J. Taylor)  
(R. McQueen)

Argonne National Lab, Libraries Dept, Rept Sect, Bldg 23, RI-CE-125,  
9700 S. Cass Ave, Argonne, IL 60440  
USAE, Albuquerque Ops Ofc ATTN: Wpn Sys Safety Br, P. O. Box 5400,  
Albuquerque, NM 87111  
Courant Institute of Mathematical Sciences, AEC Computing and Applied  
Math Cen (Tech Lib), 251 Mercer St, New York, NY 10013  
OTHER

Langley Research Cen (NASA), ATTN: Assoc Dir, Langley Sta, Hampton,  
Va 23665

National Bureau of Standards, Radiological Equipment Sect, Wash, DC  
20234

## DISTRIBUTION (cont'd)

No. cys

1 Director, National Bureau of Standards, Central Radio Propagation  
Laboratory, Boulder, Colo 80302  
1 Manned Spacecraft Cen (NASA), ATTN: Chief, Tech Info Div, Houston,  
Tex 77001  
1 Institute for Defense Analysis, Km 2B257, Pentagon, Wash, DC 20330  
THRU: APPA  
1 Research Analysis Corp (M. I. Emerson, Librarian), McLean, Va 22101  
1 MIT, Lincoln Laboratory (Doc Lib), P. O. Box 73, Lexington, Mass  
02173  
Aerospace Corp P. O. Box 95085, Los Angeles, Calif 90045  
(W. Barry)  
(S. Siegel)  
(R. M. Cooper)  
(J. McClelland)  
TRW Systems, Eng Mechs Lab, One Space Park, R 1/1004, Redondo Beach,  
Calif 90278  
(F. B. Fay, Staff Eng)  
(Dr. J. Slaughter)  
Aerospace Corp, San Bernardino Open, P. O. Box 1308, San Bernardino,  
Calif 92402  
(Ali M. Aqvi)  
(J. Kehler)  
(D. Glenn)  
(C. Francis)  
Johns Hopkins University, Applied Physics Laboratory, 8621 Georgia  
Ave, Silver Spring, Md 20910  
AVCO Corp, Rad Div, 201 Lowell St, W. M. Washington, Mass 01887  
(Chief Librarian)  
(M. C. Atkins)  
(W. L. Bade)

EGG, ATTN: A. R. Clyde, 690 Sunset Rd, Las Vegas, Nev 89101  
Northrop Corp, ATTN: D. V. Keller, Applied Rad Dept, 2323 Teller  
Rd, Newbury Park, Calif 91320  
MIT Lincoln Laboratory, P. O. Box 73, Lexington, Mass 02173  
(Dr. J. H. Pannell)  
(Dr. E. Pike)

E. H. Plesset Assoc, Inc., 2444 Wilshire Blvd, Santa Monica, Calif  
90403

## DISTRIBUTION (Cont'd)

No. cys No. s

Battelle Memorial Institute, 505 King Ave, Columbus, Ohio 43201  
(Library)  
(Mr. R. Castle)

The Dikewood Corp, 4805 Menaul Blvd NE, Albuquerque, NM 87110

General Atomic Div, General Dynamics Corp, P. O. Box 608, San Diego, Calif 92112  
(Library)  
(J. Nuance)

General Electric Co, MSN, ATTN: Dr. F. A. Lucy, Rm 99505, P. O. Box 8555, Philadelphia, Pa 19101

General Electric Co, ATTN: A. A. Sinigaglia, Dev Engr, Spacecraft Dept, King of Prussia Park, P. O. Box 8661, CCSP 7, Rm 7246, Philadelphia, Pa 19105

DASA Information and Analysis Cen, TEMPO, General Electric Co, 816 State St, Santa Barbara, Calif 93102

MIT, Div of Industrial Cooperation, 77 Massachusetts Ave, Cambridge, Mass 02139

Paul Weidinger, CE, 777 Third Ave, New York, NY 10017

University of Illinois, ATTN: Dr. A. M. Newmark, Head, Civil Engineering Dept, 1114 Civil Engineering Bldg, Urbana, Ill 61801

College Park Metallurgy Cen, US Bureau of Mines, ATTN: Library, College Park, Md 20740

University of Massachusetts, Head, Civil Engineering Dept, Amherst, Mass 01002

University of Michigan, ATTN: Prof B. G. Johnston, Dept of Civil Engineering, Ann Arbor, Mich 48104

Space Technology Laboratories, Inc., P Contr Mgt Ofc, c/o STIL, Bldg S1930, One Space Park, Redondo Beach, Calif 90278

Stanford Research Institute, ATTN: Dr. C. G. Doran, Dir, Shockwave Physics Div, Menlo Park, Calif 94025

Washington State University, Dept of Physics, Pullman, Wash 99163  
(Dr. G. R. Fowles)

(G. Duvall)

Lockheed Missiles and Space Co, ATTN: Dr. D. Miffat, 1111 Lockheed Way, Sunnyvale, Calif 94086

Bell Telephone Laboratories, Inc., Whippny, NJ 07981  
(J. W. Foss)  
(Dr. W. Troutman)

## DISTRIBUTION (cont'd)

No. cys No. s

Kaman Nuclear, Garden of the Gods, Road, Colorado Springs, Colo 80907  
(F. H. Shelton)  
(D. Williams)

(L. Battelle)  
(Dr. A. P. Bridges)

Boeing Aerospace Div, P. O. Box 3985, Seattle, Wash 98124  
(G. Butler)  
(R. Young)  
(J. Penning)

Douglas Act Corp, 3000 Ocean Park Blvd, Santa Monica, Calif 90405  
(B. Battelle)  
(J. Peck)  
(D. H. Dent)  
(R. Reck)  
(B. Love)

Aerojet General Corp, ATTN: Dr. Fischer, 1171 W. Adruff Ave, Downey, Calif 90241  
(Dr. C. Godfrey)

Physics International Corp, 2700 Merced St, San Leandro, Calif 94577  
(Dr. W. Bierman)  
(Dr. C. Godfrey)

Shock Hydrodynamics, Inc., 15010 Ventura Blvd, Sherman Oaks, Calif 91403  
(R. Bjork)

(H. Wagner)

General Motors Mfg Dev Corp, General Motors Tech Cen, Warren, Mich 48090  
(C. Madden)  
(C. Jones)  
(W. Isbell)

Union Carbide Corp, P. O. Box 278, Tarrytown, NY 10017  
(Dr. C. Zarby)  
(Dr. F. LoSacco)

Applied Theory, Inc., ATTN: Dr. J. Trulio, 1935 Cotner Ave, Los Angeles, Calif 90025  
McDonnell Afcft Corp, ATTN: L. Laczini, P. O. Box 574, St. Louis, Mo 63116

Aeroneutronic, ATTN: Dr. R. Grandey, Ford Rd, Newport Beach, Calif 92660



UNCLASSIFIED  
Security Classification

10	KEY WORDS	LINK A		LINK B		LINK C	
		ROLE	WT	ROLE	WT	ROLE	WT
Equation of state							
Hugoniot							
Gruneisen parameter							
Shock transport							

INSTRUCTIONS

1. ORIGINATING ACTIVITY: Enter the name and address of the contractor, subcontractor, Grantee, Department of Defense activity or other organization (corporate author) issuing the report.

2a. REPORT SECURITY CLASSIFICATION: Enter the overall security classification of the report. Indicate whether "Restricted Data" is included. Marking is to be in accordance with appropriate security regulations.

2b. GROUP: Automatic downgrading is specified in DoD Directive 5200.10 and Army Forces Leader of Normal, Executive, Group 3 and Group 1. Also, when applicable, show that markings have been used (for Group 3 and Group 1) and marked.

3. REPORT TITLE: Enter the complete report title in all capital letters. Titles in all cases should be unclassified. If a sensitive title cannot be selected, without classification, show the classification in all capitals in parentheses immediately following the title.

4. DESCRIPTIVE NOTES: If appropriate, enter the type of report, e.g., interim, progress, summary, annual, or final. Give the inclusive date when a specific reporting period is covered.

5. AUTHOR(S): Enter the name(s) of author(s) as shown on or in the report. Enter last name, first name, middle initial. If military, show rank and branch of service. The name of the principal author is an absolute minimum requirement.

6. REPORT DATE: Enter the date of the report, as day, month, year, or date, year. If more than one, list, separated on the report, the date of publication.

7a. TOTAL NUMBER OF PAGES: The total page count should follow normal pagination procedures, i.e., enter the number of pages containing information.

7b. NUMBER OF REFERENCES: Enter the total number of references cited in the report.

8a. CONTRACT OR GRANT NUMBER: If appropriate, enter the applicable number of the contract or grant under which the report was written.

8b. Dr. & M. PROJECT NUMBER: Enter the appropriate military department identification, such as project number, subproject number, system number, task number, etc.

9a. ORIGINATOR'S REPORT NUMBER(S): Enter the official report number by which the document will be identified and controlled by the originating activity. This number must be unique to this report.

9b. OTHER REPORT NUMBER(S): If the report has been assigned any other report numbers (either by the originator or by the sponsor), also enter the numbers.

10. AVAILABILITY/LIMITATION NOTICES: Enter any limitations or further dissemination of the report, other than those

(1) "Qualified" means there may obtain copies of this report from DDCI.

(2) "Development management and dissemination of this report by DDCI is not authorized."

(3) "U. S. Government agencies may obtain copies of this report directly from DDCI. Other qualified DDCI users shall request through:

(4) "U. S. military agencies may obtain copies of this report directly from DDCI. Other qualified users shall request through:

11. SUPPLEMENTARY NOTES: Use for additional explanatory notes.

12. SPONSORING MILITARY ACTIVITY: Enter the name of the departmental project officer or task manager sponsoring (paying for) the research and development activities.

13. ABSTRACT: Enter an abstract giving a brief and factual summary of the document indicative of the report, even though it may also appear elsewhere in the body of the technical report. If additional space is required, a continuation sheet shall be attached.

14. KEY WORDS: Key words are technically meaningful terms or short phrases that characterize a report and may be used as an indicator of the subject matter or cataloging the report. There is no limitation on the length of the abstract. However, the suggested length is from 150 to 225 words.

APSC (NAPC) 144

UNCLASSIFIED  
Security Classification

Unclassified

Unclassified

When this document is no longer required  
by your activity, DESTROY IT in accordance  
with applicable security regulations.  
DO NOT RETURN IT TO DDC

When this document is no longer required  
by your activity, DESTROY IT in accordance  
with applicable security regulations.  
DO NOT RETURN IT TO DDC

Unclassified

Unclassified